Abstract

The invention relates to a seal arrangement for a gas turbine.

The seal arrangement is used for sealing a gap (19) between radially internally located ends (18) of guide vanes (16) of a guide vane ring (15) and a rotor (12), in which case the rotor (12) has at least two seal projections (25, 26) positioned at an axial distance relative to each other in circumferential direction of the rotor (12), said seal projections (25, 26) effecting a seal of the gap (19) in combination intake linings (27, 28) associated with the radially internally located ends (18) of the guide vanes (16).

In accordance with the invention, the seal projections (25, 26) are inclined or tilted in axial direction toward a side of higher pressure, whereby, in a space (29) limited by the minimum of two seal projections (25, 26) and the corresponding intake linings (27, 28), at least one recirculation structure (30) is provided, and whereby the recirculation structure, or each recirculation structure (30), is oriented toward the side of the higher pressure.

(Fig. 1)